

IN THE CLAIMS

1. (Currently amended) A method for fabricating a forged titanium-alloy article, comprising the steps of

providing a workpiece made of an alpha-beta titanium alloy having a nominal composition in weight percent of 6 percent aluminum, 4 percent vanadium, 0.2 percent oxygen, balance titanium and impurities, wherein the titanium alloy has a beta-transus temperature; thereafter

forging the workpiece to make a forged gas turbine engine component, wherein the forged gas turbine engine component has a thick portion thereof with a section thickness greater than 2-1/4 inches; thereafter

rough machining the forged gas turbine engine component; thereafter

heat treating the forged gas turbine engine component by

solution heat treating the machined, forged gas turbine engine component at a temperature of from about 50°F to about 75°F below the beta-transus temperature, thereafter

water quenching the gas turbine engine component to room temperature, and thereafter

aging the gas turbine engine component at a temperature of from about 900°F to about 1000°F; and thereafter

final machining the forged gas turbine engine component in the absence of additional heat treatment steps.

2. (Withdrawn) The method of claim 1, wherein the step of providing the workpiece includes the steps of

preparing a melt of the titanium alloy, thereafter

casting the melt of the titanium alloy to form an ingot, thereafter

converting the ingot to a billet by hot working, and thereafter

cutting the billet transversely to form a mult that serves as the workpiece.

3. (Original) The method of claim 1, wherein the step of forging the workpiece includes the step of

forging the workpiece to make the forged gas turbine engine component selected from the group consisting of a compressor disk, a fan disk, and a gas turbine engine mount.

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4. (Original) The method of claim 1, wherein the step of forging the workpiece includes the step of

forging the workpiece to make a forged compressor disk or a forged fan disk.

5. (Original) The method of claim 1, wherein the step of solution heat treating includes the step of

solution heat treating the forged gas turbine engine component for a time of from about 45 minutes to about 75 minutes.

6. (Original) The method of claim 1, wherein the step of water quenching is initiated within about 20 seconds of completing the step of solution heat treating.

7. (Previously presented) The method of claim 1, wherein the step of aging includes the step of

aging the forged gas turbine engine component for a time of at least about 4 hours.

8. (Original) The method of claim 1, including an additional step, after the step of forging the workpiece and before the step of heat treating, of

ultrasonically inspecting the forged gas turbine engine component.

9. (Original) The method of claim 1, including an additional step, after the step of forging the workpiece and before the step of final machining, of

ultrasonically inspecting the forged gas turbine engine component.

10. (Currently amended) A method for fabricating a forged titanium-alloy article comprising the steps of

providing a workpiece made of an alpha-beta titanium alloy having a nominal composition in weight percent of 6 percent aluminum, 4 percent vanadium, 0.2 percent oxygen, balance titanium and impurities, wherein the titanium alloy has a beta-transus temperature; thereafter

forging the workpiece to make a forged gas turbine engine component, wherein the forged gas turbine engine component has a thick portion thereof with a section thickness greater than 2-1/4 inches; thereafter

rough machining the machined, forged gas turbine engine component; thereafter

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heat treating the forged gas turbine engine component by
solution heat treating the forged gas turbine engine component at a temperature of
from about 50°F to about 75°F below the beta-transus temperature, thereafter
water quenching the gas turbine engine component to room temperature, and
thereafter
aging the gas turbine engine component at a temperature of from about 900°F to
about 1000°F; and thereafter
final machining the gas turbine engine component in the absence of additional heat
treatment steps, wherein the thick portion has a 0.2 percent yield strength of from about 120 ksi
to about 140 ksi at its centerline, and a 0.2 percent yield strength of from about 160 ksi to about
175 ksi at a location about 1/2 inch below a surface thereof.

11. (Withdrawn) The method of claim 10, wherein the step of providing the
workpiece includes the steps of
preparing a melt of the titanium alloy, thereafter
casting the melt of the titanium alloy to form an ingot, thereafter
converting the ingot to a billet by hot working, and thereafter
cutting the billet transversely to form a mult that serves as the workpiece.

12. (Original) The method of claim 10, wherein the step of forging the workpiece
includes the step of
forging the workpiece to make the forged gas turbine engine component selected from
the group consisting of a compressor disk, a fan disk, and a gas turbine engine mount.

13. (Original) The method of claim 10, wherein the step of forging the workpiece
includes the step of
forging the workpiece to make a forged compressor disk or a forged fan disk.

14. (Original) The method of claim 10, wherein the step of solution heat treating
includes the step of
solution heat treating the forged gas turbine engine component for a time of from about
45 minutes to about 75 minutes.

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15. (Original) The method of claim 10, wherein the step of water quenching is initiated within about 20 seconds of completing the step of solution heat treating.

16. (Original) The method of claim 10, wherein the step of aging includes the step of
aging the forged gas turbine engine component for a time of at least about 4 hours.

17. (Original) The method of claim 10, including an additional step, after the step of forging the workpiece and before the step of heat treating, of
ultrasonically inspecting the forged gas turbine engine component.

18. (Original) The method of claim 10, including an additional step, after the step of heat treating and before the step of final machining, of
ultrasonically inspecting the forged gas turbine engine component.

19-20. (Cancelled)

21. (Previously presented) The method of claim 1, wherein the step of final machining includes the step of
removing the alpha-case at a surface of the gas turbine engine component.

22. (Previously presented) The method of claim 10, wherein the step of final machining includes the step of
removing the alpha-case at the surface of the gas turbine engine component.